

Interoperability for satellite tracking aids: the Satellite Retroreflector Standards

Standards for Dark & Quiet Skies - Lumi Space - CleanSpace Industry Days

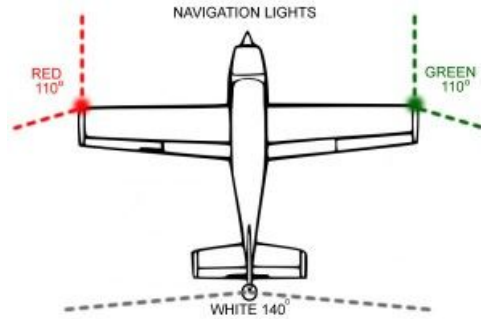
Hira Virdee, David Gooding, Ewan Schafer, Liam Pieters
09-10-2024



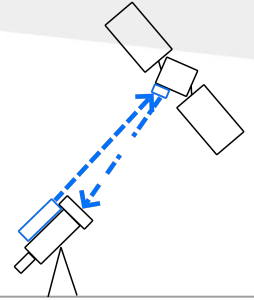
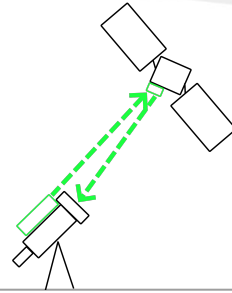
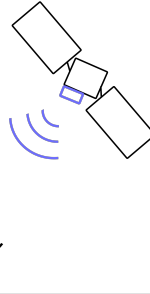
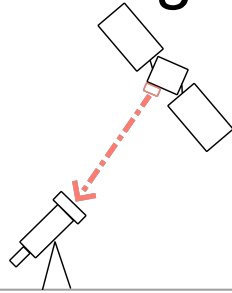
Why tracking aids?



Why standardise?



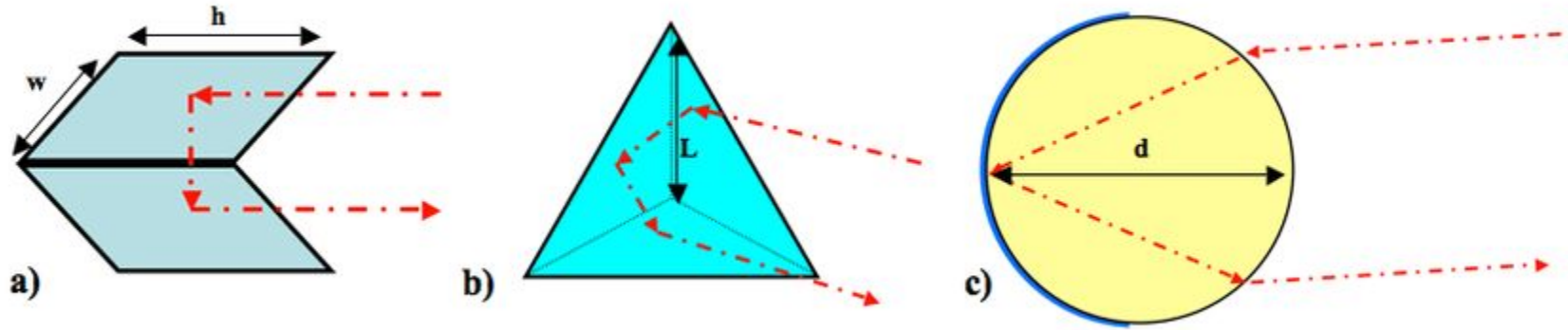
Types of tracking aid



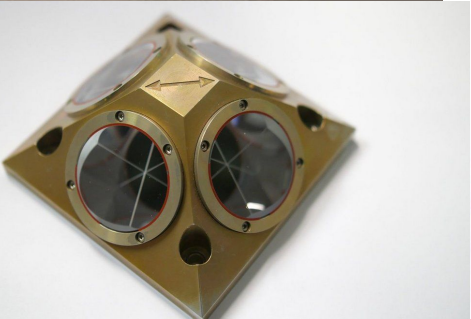
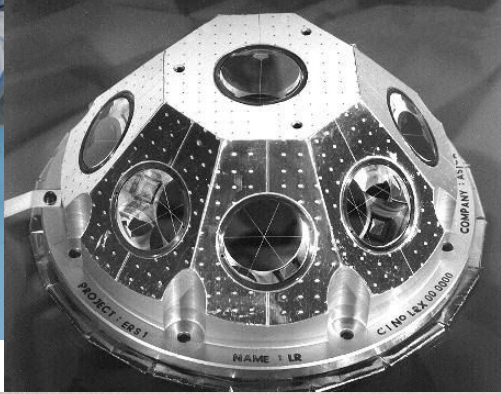
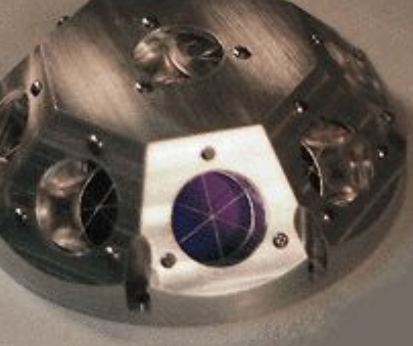
	LED beacon	Radio transmitter	Retroreflector	Modulated retroreflector
Integration + regulatory hurdle	Medium 😐	High 😞	Low 😊	Medium 😐
Accuracy	Low 😞	Medium/high 😊	High 😊	High 😊
Power need	Low/medium 😊	Medium 😐	Zero 😊	Low 😊
Resilience	Medium 😐	Low 😞	High 😊	Medium 😐
Dark/quiet skies	Poor 😞	Poor 😞	Good 😊	Good 😊



Retroreflectors



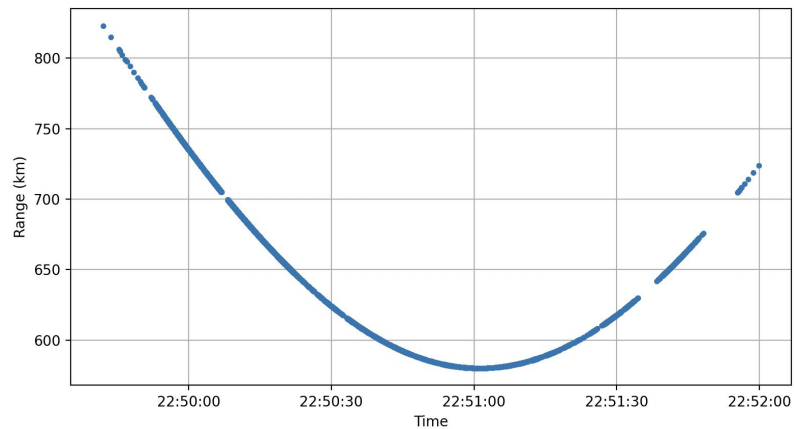
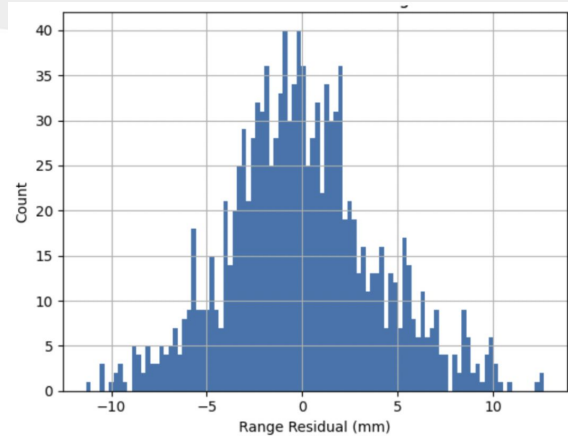
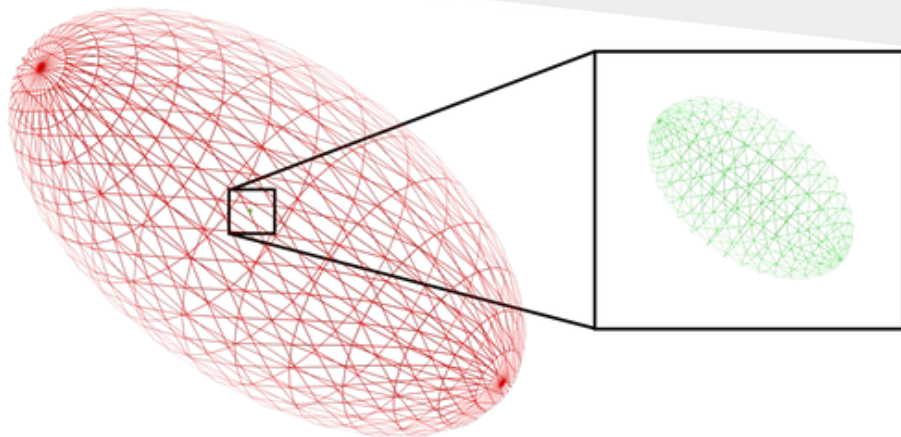
Williams, Gateman, Goyette and Giles, "Radar cross section measurements of frequency selective terahertz retroreflectors"



Lumi

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Retroreflector standards and dependencies

Dependencies for standards:

- Space segment
 - Altitude/orbit
 - Placement
 - Corner cubes
 - Quantity
 - Size
 - Arrangement
 - Quality
 - Mass

Not dependencies:

- Bus size/mass
- Power availability
- Communications

Not for these standards

- Ground segment
 - Laser power
 - Pulse width
 - Wavelength
 - Location



Types that exist today

- Geodesy-grade:
 - mm-level accuracy
 - Minimise target signature effect
 - Coincident optical centres of the cubes
 - Spherically symmetrical satellites
 - Recessed / narrow acceptance angle
- SSA-grade:
 - cm-level
 - Increase detectability
 - Improve trackability
 - Increase
 - 10x cheaper
 - 5x lighter

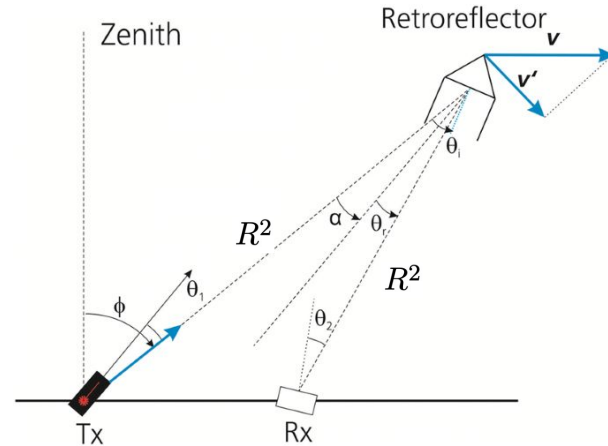


Satellite Retroreflector Standards

Optimal cross-section by altitude

- Quartic dependence
 - R^2 on beam divergence out
 - R^2 on beam divergence back from satellite
- Retroreflector increases the “optical size” of the target
- Higher altitude -> Larger cross-section needed

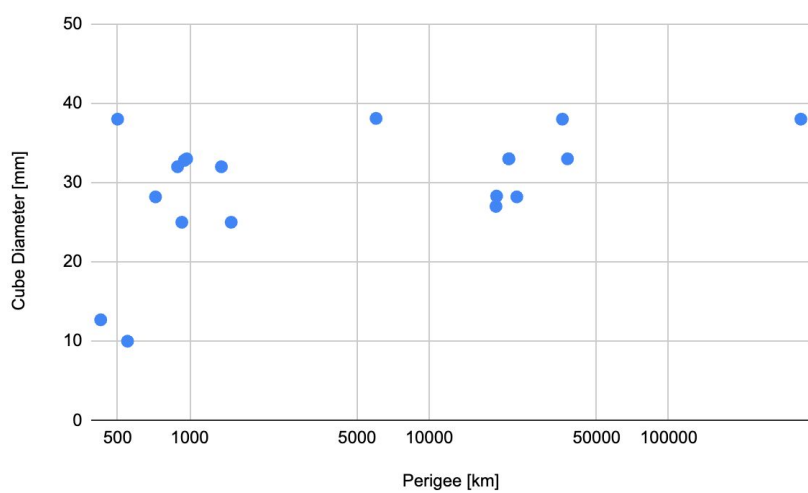
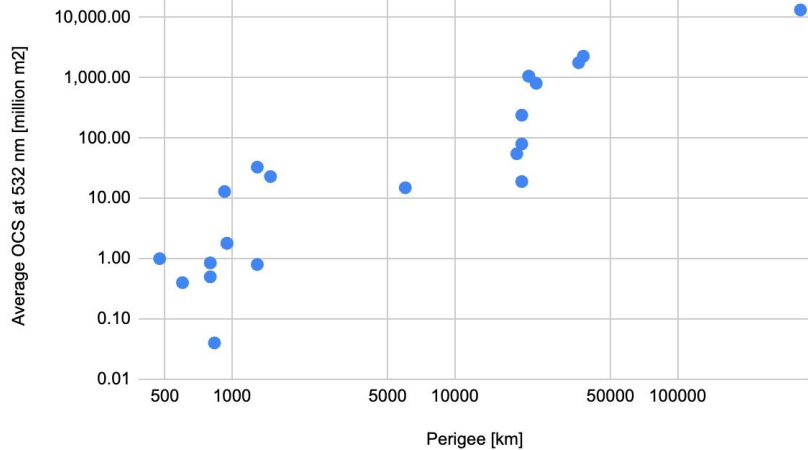
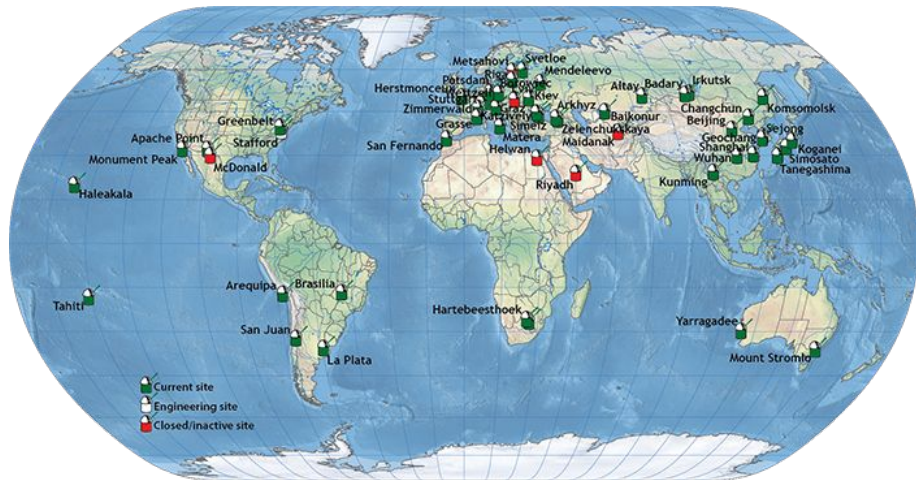
$$N_{pe} = N_t \cdot G_t \cdot \left(\frac{1}{4 \cdot \pi \cdot R^2} \right)^2 \cdot \sigma \cdot A_r \cdot \tau_r \cdot \tau_t \cdot \eta_q \cdot \tau_{filter} \cdot \tau_{atm}^2 \cdot \tau_c^2$$



False positive alert

Covariance matrix with	
$20m \times 150m \times 30m$ at TCA	$10m \times 10m \times 10m$ at TCA
An action threshold of 10^{-4} would have to be applied to reduce the collision risk by $> 90\%$	For the same risk reduction of $> 90\%$ an action threshold of 10^{-2} will be sufficient
This leads to about 2 annual manoeuvres per spacecraft on average	This leads to 0.025 annual manoeuvres per spacecraft on average
The false alert rate is at 99.9%	The false alert rate is at 10%

Setty, Flohrer & Krag,
SLR FOR SPACE DEBRIS MONITORING: AN ANALYSIS ON
REQUIREMENTS AND ACHIEVABLE ORBIT IMPROVEMENT



- Existing station compatibility / qualification
- Retros designed to have optical cross-sections far larger than necessary: Large, heavy & expensive.
- Few stations validate link performance
- Comprehensive ground-station performance validation by Tristan Meyer, at DLR Institute of Technical Physics in Stuttgart.
- Network-wide effort needed
- Critical for next-generation SLR stations

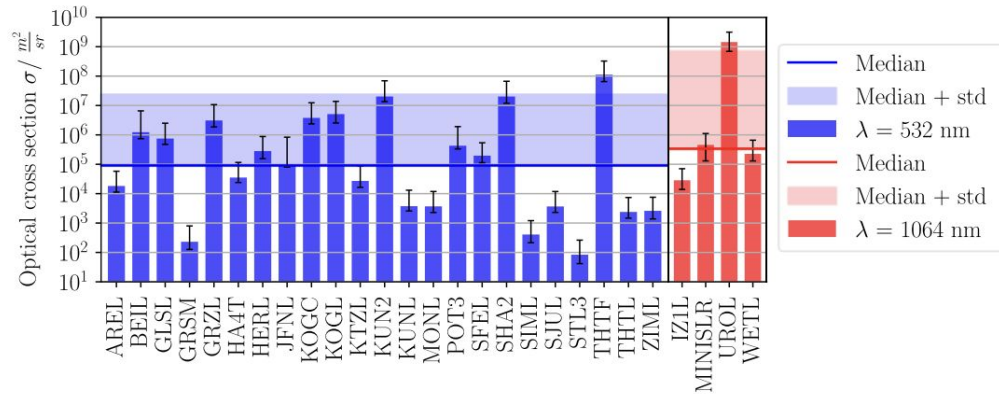
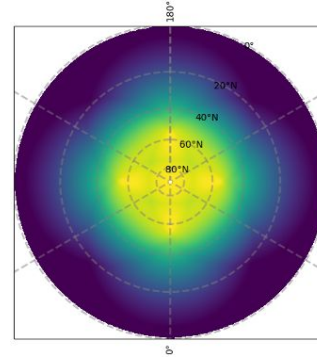


Figure from Meyer, Tristan, et al. "SLR link budget and retroreflector optical cross section evaluation." (2022).

Scoring

- Scoring outline (Issue 1):
 - Optical Cross Section
 - Far-Field Diffraction
 - Visibility / placement
 - Spectral Compatibility
- Attitude Determination and other applications:
 - Planned for issue 2
- Identification:
 - Planned for issue 3



Visibility score

- Total solid angle of acceptance where the OCS above threshold
 - Does the spacecraft operate in a fixed attitude mode?
 - Fraction of earth disk visible to the CCR
- Shouldn't only consider operational attitude - End of Life
- Lumi policy:
 - We do not track missions with a visibility score < 0.25
 - Or it's going to cost more due to incompatible passes



Velocity Aberration

The returning beam is dragged in the direction of relative velocity by a small angle.

$$\alpha \approx 2 v/c$$

Depends on the component of the relative velocity tangential to the line-of-sight.

E.g. for circular orbits: greatest at zenith, lowest at the horizon.

Most significant in LEO where orbital velocities are highest.

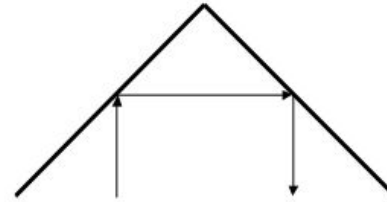


Figure 1a. Ray entering the left half.

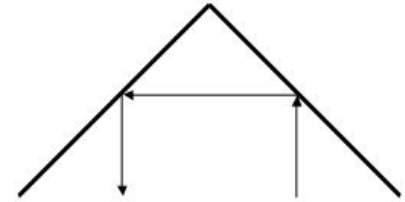


Figure 1b. Ray entering the right half.

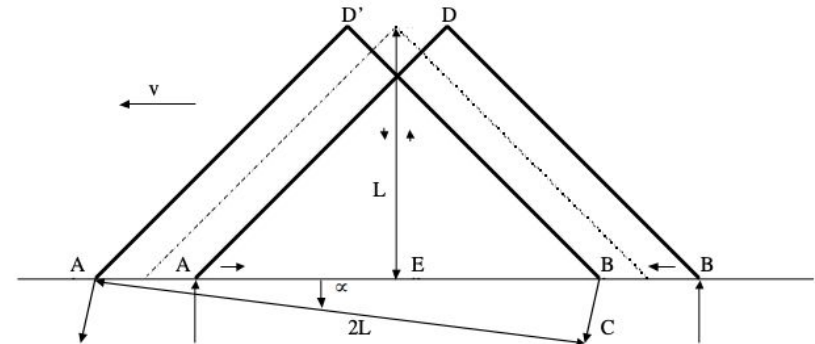


Figure 4. Moving retroreflector

Figures from David A. Arnold "Velocity Aberration"

FFDP Score:

Min / max velocity aberration defines an annulus of possible observation geometries

Assign a score based on the fraction of the energy in the diffraction pattern that falls in this annulus.

Encourages use of smaller diameter cubes at lower altitude.

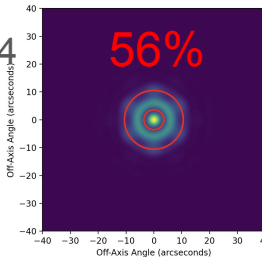
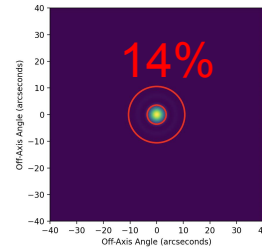
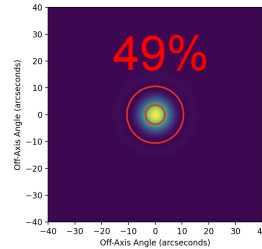
Spoiling large diameter cubes is a good strategy if the ranging wavelength is known, but can adversely affect stations that are not in the designed wavelength range.

12.7 mm
CCR in
400 km
orbit

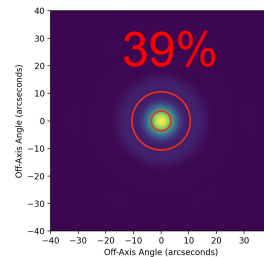
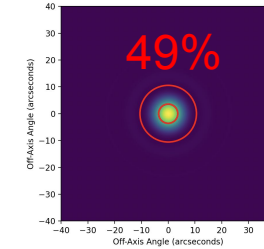
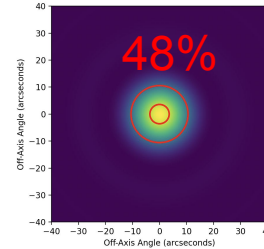
25.4 mm
CCR in
400 km
orbit

Spoilt 25.4
mm CCR
in 400 km
orbit

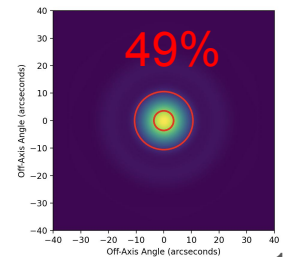
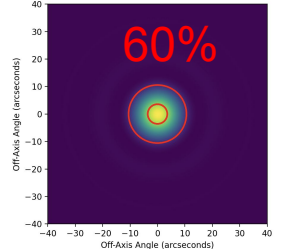
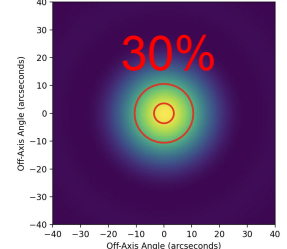
532 nm



1064 nm



1550 nm



Spectral Compatibility Score

Some LLRs use anti-reflection coatings e.g. Galileo-2XX

For the sake of improving the performance by a few % in the design wavelength, you make it much worse in others.

Discourage use of front-side AR coatings.

Conclusion

- Tracking aids are needed
 - Retroreflectors are easiest
 - Standards are important
 - Reach out!
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